



STRUCTURAL ANALYZING IN GEAR BOX COVER USING NX-NASTRAN

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Keywords: Displacement, Path length, Stress, Max shear and deflection.

Abstract

At first we used prototype model to analyze the component performance under varying load condition. Before manufacturing we found the capacity of component with the help of prototype. But it had some demerits like as more lead-time and redesign cost. Due to this we introduce new analyzing tools Nx-Nastran and Ansys. Here we selected Nx-Nastran to analyze the gear box cover. NX Nastran is a powerful, general purpose Finite Element Analysis (FEA) tool with an integrated graphical user interface and model, which is used to analyze linear and nonlinear stress, dynamics, and heat transfer characteristics of structures and mechanical components. It represents the latest in FEA technology with some of the fastest solvers on the market along with accurate solutions that have been trusted for over 20 years by companies in all industries. NX Nastran is available on a wide variety of platforms including 32-bit and 64-bit Windows and Linux operating systems.

NX Nastran (NASA Structural analysis) is a series of commercial software products originally developed under a NASA contract in the late 1960s by MSC Software Corporation using FORTRAN programming language. It uses the Finite Element Method which discretizes geometry into small elements and solves large sparse matrices using linear algebra to find quantities like displacement and stress in order to design structures. It became the industry standard program in part due to MSC buying competitors and in incorporating their advances into their products. After an antitrust settlement with the FTC in 2002, their source code was released to various organizations. Alternative Nastran versions were soon created including NEi-Nastran and NX Nastran.

Introduction

Nx-Nastran is Siemens product lifecycle management software inc. Parts of the UG/knowledge fusion software has been provided by Heidi Corporation. This product includes the international components for Unicode software, provided by international business machine co operation and others .We used Nx-Nastran to check the performance of engine cover under varying load condition in two stages.

Stage-1

In this stage we checked the performance of gear box cover under varying load condition without ribs. GBC's performance was graphed under varying load condition. X - Axis represents path length and Y - Axis explains displacement of gear box cover under varying load conditions. We attached the simulation results of gearbox cover with graph.

Gbc without rib's





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Simulation Report

Stage Solution 1

Material Name	Material Category	Material Type	Source	Category Metal
Iron_Cast_G40	METAL	Isotropic	Library	Mass Density (RHO)7.15e-006 kg/mm ³ Young's Modulus (E)1.4e+008 MN/MM ² (KPA) Poisson's .25 Yield Strength345000 MN/MM ² (KPA) Ultimate Tensile Strength570000 MN/MM ² (KPA) Fatigue Strength Coefficient645000 MN/MM ² (KPA) Fatigue Strength Exponent-0.078 Fatigue Ductility Coefficient0.037 Fatigue Ductility Exponent-0.457 Initial Strain0.02 mm/mm Hardening Exponent0.21 Strength Coefficient975.912 N/mm ² (MPA) R0 1.8 R451.8 R901.8

Modeling Objects ummary

Modeling Object Label	Modeling Object Name	Modeling Object Type
1	Bulk Data Echo Request1	Bulk Data Echo Request
2	Structural Output Requests1	Structural Output Requests

Meshes

Total number of meshes in the part	1
Total number of elements in the part	25231
Total number of nodes in the part	46328
Total number of Tetra10 elements in the part	25231

Mesh	Element Family	Elements	Nodes
Solid(1) : PSOLID1 , Iron_Cast_G40 (Material inherited)			
3d_mesh(1)	Tetra10	25231	46328

Solution steps

Number of steps in the solution: 1

Step Name	Number of referenced loads	Loads		
Sub case - Static Loads 1	1			
		Pressure(1)	Type	Pressure - Normal pressure on 2D elements or 3D element faces
			Solver Card Name	PLOAD4
			Layer	1
			Applied to	1 Polygon Face
			Pressure	1500 N/mm ² (MPA)
			Method	Constant



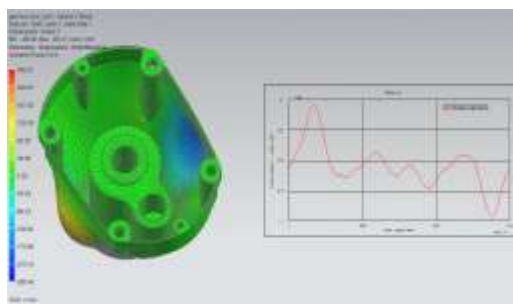
Constraints

Step Name	Number of referenced constraints	Constraints		
Subcase - Static Loads 1	1			
		Fixed(1)	Type	Fixed - Fixed
			Solver Card Name	SPC
			Layer	1
			Applied to	1 Polygon Face
			Description	

**Results Summary
Structural results**

*Coordinate System: Absolute Rectangular
Number of load cases: 1*

	Displacement (mm)				Stress (MN/MM ² (KPA))			
	X	Y	Z	Magnitude	Von-Messes	Min Principal	Max Principal	Max Shear
Static Step 1								
Max	2.556e+001	2.522e+002	7.188e+001	2.650e+002	4.202e+008	1.207e+008	5.284e+008	2.188e+008
Min	- 4.034e+001	- 2.555e+002	- 2.891e+001	0.000e+000	5.818e+004	-2.833e+008	-4.156e+007	3.348e+004



Graph result of gbc with out ribs

Stage-2

In this stage we attached the ribs in gear box cover. Here we got high performance rate. Because deformation was less when comparing with first stage. Above simulation result clearly explained about the deformation of GBC.



Graph of gbc with rib's



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Simulation Report

Stage Solution 2

Material Name	Material Category	Material Type	Source	Category Metal
Iron_Cast_G40	METAL	Isotropic	Library	Mass Density (RHO)7.15e-006 kg/mm ³ Young's Modulus (E)1.4e+008 MN/MM ² (KPA) Poisson's0.25 Yield Strength345000 MN/MM ² (KPA) Ultimate Tensile Strength570000 MN/MM ² (KPA) Fatigue StrengthCoefficient645000mN/mm ² (KPA) Fatigue Strength Exponent-0.078 Fatigue Ductility Coefficient0.037 Fatigue Ductility Exponent-0.457 Initial Strain0.02 mm/mm Hardening Exponent0.21 Strength Coefficient 975.912 N/mm ² (MPa) R01.8 R451.8 R901.8

Meshes

Total number of meshes in the part:	1
Total number of elements in the part:	25231
Total number of nodes in the part:	46328
Total number of Tetra10 elements in the part:	25231

Mesh	Element Family	Elements	Nodes
Solid(1) : PSOLID1 , Iron_Cast_G40 (Material inherited)			
3d_mesh(1)	Tetra10	25231	46328

Load

Step Name	Number of referenced loads	Loads
Subcase - Static Loads 1	1	
Pressure(1)	Type	Pressure - Normal pressure on 2D elements or 3D element faces
	Solver Card Name	PLOAD4
	Layer	1
	Applied to	1 Polygon Face
	Description	
	Pressure	1500 N/mm ² (MPa)
	Method	Constant

Constraints

Step Name	Number of referenced constraints	Constraints		
Subcase - Static Loads 1	1			
		Fixed(1)	Type	Fixed - Fixed
			Solver Card Name	SPC
			Layer	1
			Applied to	1 Polygon Face
			Description	

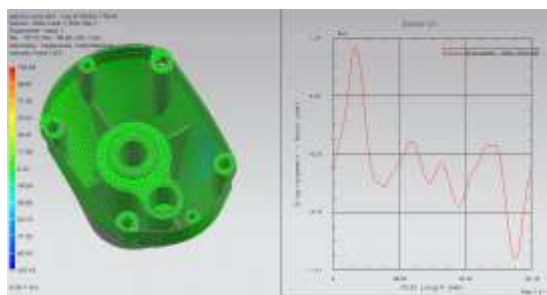


Results Summary
Structural Results

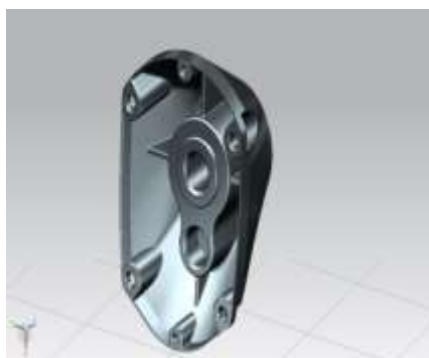
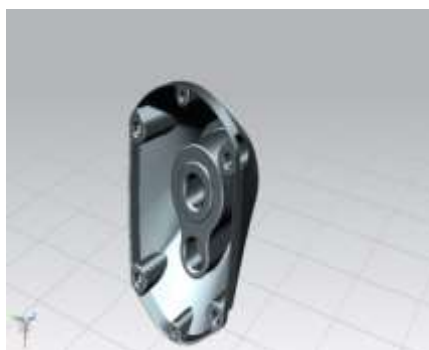
Coordinate System: Absolute Rectangular
Number of load cases: 1

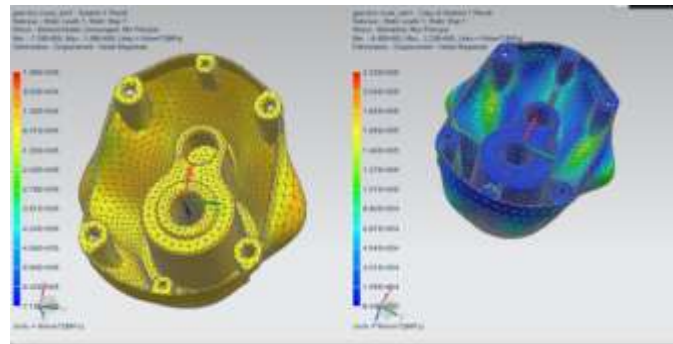
Sub case - Static Loads 1 : Number of Iterations = 1								
	Displacement (mm)				Stress (mN/mm ² (kPa))			
	X	Y	Z	Magnitude	Von-Mises	Min Principal	Max Principal	Max Shear
Static Step 1								
Max	1.709e+001	1.067e+002	3.340e+001	1.122e+002	3.434e+008	9.241e+007	4.498e+008	1.908e+008
Min	- 4.009e+001	- 1.072e+002	- 9.629e+000	0.000e+000	5.381e+004	- 1.873e+008	- 4.115e+007	3.085e+004

This graph explained about displacement and path length of GBC's with ribs. Displacement was less. It will be used for heavy load carrying application.

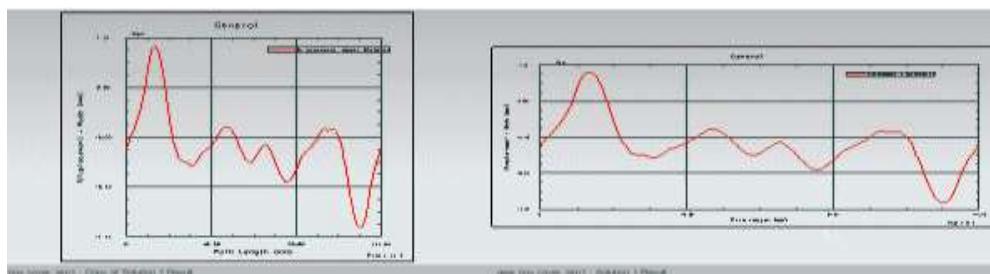


Comparison of simulation results





Above figure explains the simulation comparison of gear box cover with ribs and without ribs.



Comparison of graphs

Comparison of results
Stage Solution 1

	Displacement (mm)				Stress (mN/mm ² (kPa))			
	X	Y	Z	Magnitude	Von-Moses	Min Principal	Max Principal	Max Shear
Static Step 1								
Max	2.556e+001	2.522e+002	7.188e+001	2.650e+002	4.202e+008	1.207e+008	5.284e+008	2.188e+008
Min	-4.034e+001	-2.555e+002	-2.891e+001	0.000e+000	5.818e+004	-2.833e+008	-4.156e+007	3.348e+004

Stage Solution 2

	Displacement (mm)				Stress (MN/mm ² (kPa))			
	X	Y	Z	Magnitude	Von-Moses	Min Principal	Max Principal	Max Shear
Static Step 1								
Max	1.709e+001	1.067e+002	3.340e+001	1.122e+002	3.434e+008	9.241e+007	4.498e+008	1.908e+008
Min	-4.009e+001	-1.072e+002	-9.629e+000	0.000e+000	5.381e+004	-1.873e+008	-4.115e+007	3.085e+004

Conclusion

By using Nx Nastran the gear box cover would be analyzed with help of structural analysis, above comparison of graphs shows the gear box cover with ribs have more power to carry heavy load and also have to absorb more stresses. The displacement value of gearbox cover without rib is higher than the value of with rib gear box cover. In second stage we attached the ribs in gear box cover. So we got high performance rate. Because deformation was less when comparing with first stage. With help of Nx Nastran the displacement, path length, stress, maximum shear and deflection of gear box cover with and without ribs are found and plotted the graph



References

1. Barnes R. A., Schmid, R., Adrick, H. C., Rotor Dynamic Analysis with MSC/NASTRAN via the Important Modes Method, The 1989 MSC World Users Conf. Proc., Vol. I, Paper No. 13, March, 1989.
2. Bella., David. F., Stein hard. E., Critical Frequency Determination of a Flexible Rotating Structure Attached to a Flexible Support, Proc. of the 18th MSC Eur. Users' Conf. Paper No. 28, June, 1991.
3. Bindolino. G., Lanz M., Bassoli M., Optimal Shape of Composite Flex beams for Helicopter Rotors, MSC 1995 European Users' Conf. Proc., Italian Session, and September, 1995.
4. Cronkhite J. D., Smith. Michael R., Experiences in NASTRAN Airframe Vibration Prediction at Bell Helicopter Textron, American Helicopter Soc. Dynamics Specialists Mtg., Section 6, Vibrations Session I, Paper No. 1, November 1989.
5. Drago., Raymond J., Margasahayam., Ravi., Stress Analysis of Planet Gears with Integral Bearings; 3-D Finite Element Model Development and Test Validation, The MSC 1987 World Users Conf. Proc., Vol. I, Paper No. 4, March, 1987.
6. Ferg. D., Foote. L., Korkosz. G., Straub. F., Toossi. M., Weisenburger. R., Plan, Execute, and Discuss Vibration Measurements, and Correlations to Evaluate a NASTRAN Finite Element Model of the AH-64 Helicopter Airframe, National Aeronautics and Space Administration, January, 1990, (NASA CR-181973).
7. Hodges Robert. V., Nixon Mark .W., Rehfield Lawrence .W., Comparison of Composite Rotor Blade Models: A Coupled-Beam Analysis and an MSC/NASTRAN Finite-Element Model, National Aeronautics and Space Administration, March, 1987, (NASA TM-89024).
8. Howells. R. W., Sciarra. J. J., Finite Element Analysis Using NASTRAN Applied to Helicopter Transmission Vibration/Noise Reduction, NASTRAN: Users' Exper. pp. 321-340, September, 1975, (NASA TM X-3278).
9. Lawrence Charles. Aiello Robert. A., Ernst., Michael.A., McGee., Oliver .G., A NASTRAN Primer for the Analysis of Rotating Flexible Blades, The MSC 1988 World Users Conf. Proc., Vol. I, Paper No. 32, March, 1988.
10. Melli, R.; Rispoli, F.; Sciubba, E.; Tavani, F. Structural and Thermal Analysis of Avionic Instruments for an Advanced Concept Helicopter, Proc. of the 15th MSC/NASTRAN Eur. Users' Conf., October, 1988.
11. Miranda. I., Arnait.C., Martinez., Azua.J., Ruiz de. Inclusion of Gyroscopic Effects for the Computation of Critical Speeds of Rotor Systems, Proc. of the 15th MSC/NASTRAN Eur. Users' Conf., Paper No. 8, October, 1988.
12. Morton Mark .H., Kaizoji., Allyne., Effects on Load Distribution in a Helicopter Rotor Support Structure Associated with Various Boundary Configurations, 48th Annual Forum Proc. of the American Helicopter Society, Vol. 1, pp. 629-634, 1992.
13. Nowak William., James Courtney., Dynamic Modeling and Analysis of Spinning Polygon Assemblies Using MSC/NASTRAN, The MSC 1993 World Users' Conf. Proc., Paper No. 66, May, 1993.
14. Parker., Grant. Gear Analysis Using the MSC/NASTRAN Cyclic Symmetry Approach with Enhanced Graphics, MSC/NASTRAN Users' Conf. Proc., Paper No. 5, March, 1984.
15. Rose ., Ted., Using Dynamic Optimization to Minimize Driver Response to a Tire Out-of-Balance, MSC 1994 Korea Users' Conf. Proc., and December, 1994.
16. Rotelli., Richard .L., Jordan J.R., Apostal., Three-Dimensional Stress Analysis of a Helicopter Main Rotor Hub Using Cyclic Symmetry, MSC/NASTRAN Users' Conf. Proc., Paper No. 22, March, 1983.
17. Sainsbury-Carter. J. B., Conaway, John H., NASTRAN Data Generation of Helicopter Fuselages Using Interactive Graphics, NASTRAN: Users' Exper. pp. 661-678, September, 1973, (NASA TM X-2893).
18. Smith., Michael .R. Rangacharyulu. M., Wang Bo .P., Chang .Y. K., Application of Optimization Techniques to Helicopter Structural Dynamics, AIAA/ASME/ASCE/AHS/ASC 32nd Structures, Structural Dynamics, and Materials Conf., Part 1, Paper No. 91-0924, pp. 227-237, April, 1991.
19. Vollan. A., Calculation of Coupled Rotating/Non-Rotating Structures Using NASTRAN, Proc. of the MSC/NASTRAN First Italian Users' Conf., October, 1987.
20. Wilson. H. E., Cronkhite.J. D., Static and Dynamic Helicopter Airframe Analysis with NASTRAN, NASTRAN: Users' Exper. pp. 611-620, September, 1973, (NASA TM X-2893).